

1 **AIRWORTHINESS LIMITATIONS SCHEDULE**

NOMENCLATURE	PART NUMBER 	AIRWORTHINESS LIFE
MAIN ROTOR HUB		
Yoke Assembly	412-010-101-109	700 hours
Yoke Assembly	412-010-101-123	   5000 hours
Yoke Assembly	412-010-101-137	  4500 hours
Yoke Assembly	412-010-101-139	  3000 hours
Spindle Assembly	412-010-156-105	 5000 hours
Spindle and Damper Bearing Assembly	412-010-190-101	5000 hours
Spindle and Damper Bearing Assembly	412-010-190-103	10,000 hours
Spindle and Shear Bearing Assembly	412-310-102-101	10,000 hours
Pitch Horn Assembly	412-010-149-105	5000 hours
Pitch Horn Assembly	412-010-149-111	15,000 hours
Retention Bolt	412-010-124-105	 5000 hours
Retention Bolt (Expandable)	412-010-137-103	5000 hours
Damper Bridge	412-010-104-101	5000 hours
Damper Bridge	412-010-183-101	10,000 hours
Damper Bridge	412-010-183-109	15,000 hours
Damper Bridge	412-010-185-101	10,000 hours
Damper Bridge	412-010-185-109	15,000 hours
Damper Bridge	412-018-068-101	 10,000 hours or 180 months
Damper Bridge	412-010-170-101	 10,000 hours or 180 months
Pivot Bearing	412-010-106-101	 On condition
Fitting	412-010-111-101	5000 hours
Damper Yoke Set	412-010-145-101	 On condition
Damper Yoke Set	412-310-146-103	 On condition
MAIN ROTOR DROOP RESTRAINT		
Bolt (8)	MS-21250-H05-006	 120 months
MAIN ROTOR SIMPLE PENDULUM ABSORBERS		
Bolt, Bracket (16)	EWB0420D-7-36	15,000 hours
MAIN ROTOR CONTROLS		
Rephasing Lever Assembly	412-010-403-109	1250 hours
Rephasing Lever Assembly	412-010-403-113	5000 hours
Drive Link Assembly	412-010-405-101	5000 hours
Swashplate Outer Ring	412-010-407-105	 2500 hours

NOMENCLATURE	PART NUMBER 	AIRWORTHINESS LIFE
MAIN ROTOR CONTROLS (CONT)		
Swashplate Outer Ring	412-010-407-117	10,000 hours
Pitch Link Rod End Bearing	412-010-412-101	5000 hours
Pitch Link Rod End Bearing	412-010-438-101	5000 hours
Swashplate Link Rod End Bearing	412-010-412-101	5000 hours
Swashplate Link Rod End Bearing	412-010-448-101	5000 hours
Swashplate Link Rod End Bearing	412-310-400-103	 5000 hours
Swashplate Support Assembly	412-010-409-105	 5000 hours
Swashplate Support Assembly	412-010-443-101	 5000 hours
Swashplate Support Assembly	412-010-453-101	 5000 hours
Swashplate Support Assembly	412-010-453-105	15,000 hours
Gimbal Ring Assembly	204-010-404-001	9000 hours
Gimbal Ring Assembly	212-010-416-101	9000 hours
Collective Sleeve	204-011-408-003	9000 hours
Collective Sleeve	212-011-412-101	9000 hours
Collective Lever Assembly	412-010-408-101	 10,000 hours
Collective Lever Assembly	412-010-408-105	 2500 hours
Collective Lever Assembly	412-010-464-101	20,000 hours
Collective Lever Pin	412-010-465-101	20,000 hours
MAIN ROTOR CONTROL SYSTEM BOLTS		
Pitch Horn to Pitch Link (4)	20-057-5-25D	   2500 hours
Pitch Horn to Pitch Link (4)	20-057-5-28D	 2500 hours
Pitch Link-to-Rephasing Lever (4)	20-057-6-36D	   2500 hours
Pitch Link-to-Rephasing Lever (4)	20-057-6-52D	  2500 hours
Drive Link-to-Rephasing Lever (4)	20-057-6-36D	   2500 hours
Swashplate Link-to-Rephasing Lever (3)	20-057-6-38D	   2500 hours
Swashplate Link-to-Rephasing Lever (2)	20-057-6-52D	  2500 hours
Swashplate Link-to-Rotating Ring (2)	20-057-6-36D	   2500 hours
Gimbal Ring-to-Swashplate Support (2)	204-011-463-105	   2500 hours
Gimbal Ring-to-Swashplate Inner Ring (2)	204-011-463-001	   1000 hours
Gimbal Ring-to-Swashplate Inner Ring (2)	204-011-463-109	 2500 hours
Collective Lever-to-Swashplate Support (1)	20-057-8-92D	   2500 hours
Boost Tube-to-Universal (3)	20-057-5-24D	   2500 hours
Universal-to-Boost Cylinder (3)	20-057-5-24D	   2500 hours
Boost Cylinder-to-Lower Support (3)	212-001-323-001	   2500 hours
Rephasing Lever-to-Hub (4)	20-057-6-82D	  1250 hours
Rephasing Lever-to-Hub (4)	20-057-8-84D	   2500 hours

NOMENCLATURE	PART NUMBER 	AIRWORTHINESS LIFE
MAIN ROTOR SUPPLEMENTAL CONTROL SYSTEM BOLTS 		
Swashplate Inner Ring-to-R.H. Cyclic Boost (1)	20-057-5-24D	2500 hours
Swashplate Inner Ring-to-L.H. Cyclic Boost (1)	20-057-5-24D	2500 hours
Collective Lever-to-Collective Boost (1)	20-057-5-24D	2500 hours
PROPULSION AND DRIVE SYSTEM		
Planetary Spider	204-040-785-003	 On condition
Planetary Spider	412-040-785-101	2500 hours
Planetary Spider	412-040-785-103	 On condition
Main Rotor Mast Assembly	412-040-101-105	   10,000 hours or 80,000 RIN
Main Rotor Mast Assembly (used on helicopters with Mast Torque Systems)	412-040-101-121	   10,000 hours or 60,000 RIN
Main Rotor Mast Assembly	412-040-101-127	   10,000 hours or 80,000 RIN
Main Rotor Mast Assembly (used on helicopters with Mast Torque Systems)	412-040-101-129	   10,000 hours or 60,000 RIN
Main Rotor Mast Assembly (used on helicopters with Mast Torque Systems and with BHT-412-SI-62, Slope Landing Kit)	412-040-101-135	   5000 hours or 60,000 RIN
Main Rotor Mast Assembly	412-040-114-101	   2,500 hours or 25,000 RIN
Main Rotor Mast Assembly (used on helicopters with Mast Torque Systems and with BHT-412-SI-62, Slope Landing Kit)	412-704-012-105	   5000 hours or 60,000 RIN
Cap Retention	412-010-161-101	10,000 hours
Cap Retention	412-010-171-101	10,000 hours
Cap Retention	412-010-171-103	 10,000 hours
Mast Cap	412-010-171-105	5,000 hours
Cone	412-010-165-101	 10,000 hours
Cone	412-010-165-103	5,000 hours
Drive Pin	412-010-166-101	 10,000 hours
Drive Pin	412-010-166-103	5,000 hours
Upper Cone Seat	412-010-164-101	10,000 hours
Upper Cone Seat	412-010-174-101	10,000 hours
Upper Cone Seat	412-010-186-101	10,000 hours
Upper Cone Seat	412-010-186-107	5,000 hours
Upper Cone Seat	412-010-286-101	2,500 hours
Splined Plate Assembly	412-010-177-101	   Life limited (see notes)
Splined Plate Assembly	412-010-177-105	  10,000 hours or 60,000 RIN

NOMENCLATURE	PART NUMBER 	AIRWORTHINESS LIFE
PROPULSION AND DRIVE SYSTEM (CONT)		
Splined Plate Assembly	412-010-177-109	  10,000 hours or 80,000 RIN
Splined Plate Assembly	412-010-177-121	  5,000 hours or 60,000 RIN
Splined Plate Assembly	412-010-177-113	  10,000 hours or 60,000 RIN
Splined Plate Assembly	412-010-277-101	  2,500 hours or 25,000 RIN
Splined Plate Assembly	412-010-167-105	   Life limited (see notes)
Cone	412-010-169-103	10,000 hours
Cone	412-010-179-101	10,000 hours
Cone	412-010-179-105	 10,000 hours
Cone	412-010-179-107	5,000 hours
Lower Cone Seat	412-010-168-105	 10,000 hours
Lower Cone Seat	412-010-178-101	 10,000 hours
Lower Cone Seat	412-010-056-101	 2,500 hours
Lower Cone Seat	412-018-056-103	 10,000 hours
Lower Cone Seat	412-018-056-109	 5,000 hours
ELEVATOR AND CONTROLS		
Horn Assembly	205-001-914-103	 On condition
AIRFRAME		
Upper Aft Beam Cap	212-030-191-001	 On condition
TAIL ROTOR INSTALLATION		
Blade Assembly	212-010-750-009	5000 hours
Blade Assembly	412-016-100-111	5000 hours
Yoke Assembly	212-011-702-001	 5000 hours
LANDING GEAR		
High Aft Crosstube Assembly	412-050-011-101	   10,000 events
High Aft Crosstube Assembly	412-050-045-107	   20,000 events
High Aft Crosstube Assembly	412-321-104	   20,000 events
High Aft Crosstube Assembly	412-321-901	    12,000 events
FIRE EXTINGUISHER CARTRIDGES		
HTL Industries P/N 13083-5	209-062-908-015	  6 years service life/ 9 years total life
HTL Industries P/N 30900400	209-062-908-019	  6 years service life/ 9 years total life
W. Kidde and Co. P/N 804943	209-062-908-013	 6 years total life
W. Kidde and Co. P/N 804944	209-062-908-017	 6 years total life
W. Kidde and Co. P/N 895408-1	209-062-908-113	 10 years total life
W. Kidde and Co. P/N 895409-1	209-062-908-115	 10 years total life

NOTES

- 1 The airworthiness life or inspection interval for any part number contained in this schedule applies to all the successive dash numbers for that component unless it is otherwise specified.
- 2 In-service life begins the day the component is entered into service. In-service life only remains in effect while the component is installed on the helicopter.
- 3 Fail safe, refer to [Chapter 5](#) for detailed 100-hour inspection.
- 4 On-condition subject to daily or 25-hour visual inspection interval (as applicable) per scheduled inspections ([Chapter 5](#)).
- 5 The bushing 412-010-222 is installed on the main rotor yoke assemblies 412-010-101-127/-129/-133/-135/-137/-139. The bushing is assigned an unlimited airworthiness life. If removed, it is not reusable.
- 6 An on-condition retirement is assigned to spider 204-040-785-003 subject to passing a magnetic particle inspection every 3100 hours per [ASB 412-91-52A](#).
- 7 An additional 4 hours must be logged for each hoist operation performed within penalty CG region ([Figure 4-1](#)). Refer to [Chapter 5](#) and [BHT-412-FM](#).
- 8 RIN (Retirement Index Number) is the retirement life based on fatigue damage from normal helicopter lifts and takeoffs. New components will begin with an accumulated RIN of zero that will be increased as lifts and takeoffs are performed. Operators must record the number of lifts and takeoffs and increase the accumulated RIN accordingly. When the maximum RIN or retirement flight hours is/are reached, whichever occurs first, the component will be removed from service. Refer to [Information Letter GEN-94-54/Information Letter GEN-03-94](#).
- 9 Retire when the part has 10,000 flight hours or when the accumulated RIN = 80,000, whichever occurs first. For manual tracking, increase the RIN count by 1 for each takeoff/lift recorded. If logging, increase RIN count by 2 for each lift recorded.
- 10 Retire when the part has 10,000 flight hours or when the accumulated RIN = 60,000, whichever occurs first. For manual tracking, increase the RIN count by 1 for each takeoff/lift recorded. If logging, increase RIN count by 2 for each lift recorded.
- 11 This splined plate assembly can be used on all 412 models. When used on 412 HP/EP, it will be vibroetched "412 HP" and will be retired when part has 10,000 flight hours or when the accumulated RIN = 60,000, whichever occurs first. For manual tracking, increase the RIN count by 1 for each takeoff/lift recorded. If logging, increase RIN count by 2 for each lift recorded (see Note 12).
- 12 This splined plate assembly can be used on all 412 models. When used on 412/412SP, it will not be vibroetched "412 HP". Retire when part has 10,000 flight hours or when the accumulated RIN = 80,000, whichever occurs first. For manual tracking, increase the RIN count by 1 for each takeoff/lift recorded. If logging, increase RIN count by 2 for each lift recorded (see Note 11).
- 13 Deleted.
- 14 Retire when the part has 5000 flight hours or when the accumulated RIN = 60,000, whichever occurs first. For manual tracking, increase the RIN count by 1 for each takeoff/lift recorded. If logging, increase RIN count by 2 for each lift recorded.
- 15 Per [ASB 412-99-97](#).
- 16 Contained in inflation valve 30905. 30908 includes squib 94455 and o-ring 568-905.
- 17 Per [ASB 412-95-89](#), the tail rotor flapping stop/yield indicator 212-011-713-103 must be used with the tail rotor yoke. The stop/yield indicator 212-011-713-103 must be inspected every 25 hours and at each incident involving a hard landing, sudden stoppage, or any other incident involving excessive tail rotor flapping. Refer to [Chapter 5](#).
- 18 Manufactured by Aeronautical Accessories, Inc. Refer to AAI Report No. ICAAA-01136 Rev. C, or subsequent, Inspection and Maintenance Instructions. Recurring visual inspection per AAI Bulletin AA-07109 and [ASB 412-08-129](#).
- 19 As per [ASB 412-98-92](#), bolt 20-057-5-25D is replaced by 20-057-5-28D.
- 20 Total life begins from the date of component manufacture and cannot be interrupted.
- 21 Calendar life begins the day the component is entered into service and cannot be interrupted.

NOTES (Cont)

-  Not applicable to subsequent dash numbers.
 -  Deleted.
 -  Replacement bearing cartridges 412-310-400-107 are on-condition. Refer to IL 412-IL-03-49.
 -  As per [ASB 412-02-110](#), bolt 204-011-463-001 is replaced by 204-011-463-109.
 -  29022968 contained in valve assembly 20022988 Rev. E and refurbishment kit 29022971.
 -  As per [ASB 412-93-73](#), bolt 412-010-124-105 is replaced by 412-010-124-109.
 -  Included in bolt kit 412-704-112-101. Bolt kit 412-704-112-101 is replaced by bolt kit 412-704-112-105.
 -  Included in bolt kit 412-704-112-103. Bolt kit 412-704-112-103 is replaced by bolt kit 412-704-112-105.
 -  Included in bolt kit 412-704-112-105.
 -  Not included in bolt kits 412-704-112-101/-103/-105.
 -  Life reduction to 2500 hours per [ASB 412-08-131](#); remove from service by 1 June 2009.
 -  Refer to [ASB 412-08-128](#).
 -  Use aft crosstube retention strap kit per [BHT-412-SI-58](#) when towing helicopters with a gross weight of 8900 pounds or greater.
 -  Swashplate support assembly 412-010-443-101 was removed from service by [ASB 412-93-71](#). All subsequent dash numbers remain in service.
 -  Swashplate support assembly 412-010-409-105 was removed from service by [ASB 412-93-71](#).
 -  Swashplate support assembly 412-010-453-101 was removed from service by [ASB 412-92-61](#). All subsequent dash numbers remain in service.
 -  500-hour penalty to be applied in accordance with AD 99-23-23 and [ASB 412-98-93](#).
 -  AD 2012-22-11 imposes on collective lever 412-010-406-101 a visual inspection at intervals not to exceed 100 hours Time-In-Service (TIS) ([ASB 412-11-148](#) Rev. A).
 -  Visual inspection at intervals not to exceed 100 hours Time-In-Service (TIS). Inspection requirements same as listed in [ASB 412-11-148](#) Rev. A.
 -  AD 2000-18-09 imposes on the upper aft beam cap 212-030-191-001 a recurring visual inspection at intervals not to exceed 100 hours Time-In-Service (TIS) ([ASB 412-00-100](#)). All aircraft with serial numbers 38001-38999 and 39101-39999 require a visual inspection in intervals not to exceed 9 hours Time-In-Service (TIS), in accordance with AD 2000-18-09 procedures.
 -  For manual tracking, increase the event count by 1 for each takeoff recorded.
 -  Part manufactured and part number assigned by AAI of Bristol, Tennessee. Retire part when event = 12,000 (event life).
 -  1000-event penalty to be applied against accumulated events of high aft crosstube assembly if used with increased gross weight kit 412-706-140 ([BHT-412-FMS-74.5](#)). 1000 event penalty to be applied on high aft crosstube assembly 412-321-901 on all aircraft serial numbers 38001-38999 and 39101-39999.
 -  On Condition indicates that there is no retirement life (i.e., unlimited). Any required inspections will be addressed by additional specific notes.
 -  Retire when the part has 2,500 flight hours or when the accumulated RIN is 25,000, whichever occurs first. For manual tracking, increase the RIN count by 1 for each takeoff/lift recorded. If logging, increase RIN count by 2 for each lift recorded.
 -  Apply a one time 7,500 hours penalty for all aircraft with serial numbers 38001-38999 and 39101-39999 on retention cap, cones, and drive pins.
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